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airfoils are
airfoil shapes
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a series of

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first digit, when multiplied by 0.15, gives the designed coefficient of lift (C L) .

Second and third digits, when divided by 2, give p , the distance of maximum camber from the leading edge (as per

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The NACA airfoil series The early NACA airfoil series, the 4-digit, 5-digit, and modified 4-/5-digit, were generated using analytical equations that

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describe the camber (curvature) of the mean-line (geometric centerline) of the airfoil section as well as the section's thickness distribution along the length of the airfoil. Later

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The NACA airfoil
section is
created from a
camber line and
a thickness
distribution
plotted
perpendicular to
the camber line.

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The equation for the camber line is split into sections either side of the point of maximum camber position (P). In order to calculate the position of the final airfoil envelope later the gradient of the camber line

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required. The
equations are:

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airfoils. During
the late 1920s

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and into the
1930s, the NACA
developed a
series of
thoroughly
tested airfoils
and devised a
numerical
designation for
each airfoil – a
four digit
number that
represented the
airfoil

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Section's
critical
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properties.

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camber 0% at 0%
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the word "NACA".
The parameters
in the numerical
code can be
entered into
equations to
precisely
generate the
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the airfoil and
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thickness forms
having the
position of
minimum pressure
of 30, 40, and
50 percent chord
and with
thickness ratios
varying from 6
percent to 15
percent.

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